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Mechanics of Aircraft structures C.T. Sun 1.5 Derive the relations given by (1.4) and (1.5). Remark: $b t V \text{ at } V y x \dots = \tau \tau : 5.1 (:) 4.1 ($ Solution: (1) Consider a very small section within the curved panel with thickness t and length L_5 .

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Dr. Sun teaches composites, elasticity, fracture mechanics, and aircraft structures. He is a Fellow of both the American Institute of Aeronautics and Astronautics (AIAA), the American Society of Mechanical Engineers (ASME), and the American Society for Composites (ASC).